

APPENDIX V

EQUATIONS

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EQUATIONS

CHAPTER 2	<u>PAGE</u>
$P = \rho g_c R T$	(Eq 2.1) 2.4
$dP_a = -\rho g dh$	(Eq 2.2) 2.4
$g_{ssl} dH = g dh$	(Eq 2.3) 2.4
$\theta = \frac{T_a}{T_{ssl}} = \left(1 - 6.8755856 \times 10^{-6} H\right)$	(Eq 2.4) 2.5
$\delta = \frac{P_a}{P_{ssl}} = \left(1 - 6.8755856 \times 10^{-6} H\right)^{5.255863}$	(Eq 2.5) 2.5
$\sigma = \frac{\rho_a}{\rho_{ssl}} = \left(1 - 6.8755856 \times 10^{-6} H\right)^{4.255863}$	(Eq 2.6) 2.6
$P_a = P_{ssl} \left(1 - 6.8755856 \times 10^{-6} H_P\right)^{5.255863}$	(Eq 2.7) 2.6
$T_a = -56.50^\circ C = 216.65^\circ K$	(Eq 2.8) 2.6
$\delta = \frac{P_a}{P_{ssl}} = 0.223358 e^{-4.80614 \times 10^{-5} (H - 36089)}$	(Eq 2.9) 2.6
$\sigma = \frac{\rho_a}{\rho_{ssl}} = 0.297069 e^{-4.80614 \times 10^{-5} (H - 36089)}$	(Eq 2.10) 2.6
$P_a = P_{ssl} \left(0.223358 e^{-4.80614 \times 10^{-5} (H_P - 36089)}\right)$	(Eq 2.11) 2.6

FIXED WING PERFORMANCE

$$V_T = \sqrt{\frac{2}{\rho_a} (P_T - P_a)} = \sqrt{\frac{2q}{\rho_a}} \quad (Eq \ 2.12) \quad 2.10$$

$$V_e = \sqrt{\frac{2q}{\rho_{ssl}}} = \sqrt{\frac{\sigma 2q}{\rho_a}} = \sqrt{\sigma} V_T \quad (Eq \ 2.13) \quad 2.11$$

$$V_{e_{Test}} = V_{e_{Std}} \quad (Eq \ 2.14) \quad 2.12$$

$$V_T^2 = \frac{2\gamma}{\gamma-1} \frac{P_a}{\rho_a} \left[\left(\frac{P_T - P_a}{P_a} + 1 \right)^{\frac{\gamma-1}{\gamma}} - 1 \right] \quad (Eq \ 2.15) \quad 2.13$$

$$V_T = \sqrt{\frac{2\gamma}{\gamma-1} \frac{P_a}{\rho_a} \left[\left(\frac{q_c}{P_a} + 1 \right)^{\frac{\gamma-1}{\gamma}} - 1 \right]} \quad (Eq \ 2.16) \quad 2.13$$

$$q_c = q \left(1 + \frac{M^2}{4} + \frac{M^4}{40} + \frac{M^6}{1600} + \dots \right) \quad (Eq \ 2.17) \quad 2.13$$

$$V_c^2 = \frac{2\gamma}{\gamma-1} \frac{P_{ssl}}{\rho_{ssl}} \left[\left(\frac{P_T - P_a}{P_{ssl}} + 1 \right)^{\frac{\gamma-1}{\gamma}} - 1 \right] \quad (Eq \ 2.18) \quad 2.14$$

$$V_c = \sqrt{\frac{2\gamma}{\gamma-1} \frac{P_{ssl}}{\rho_{ssl}} \left[\left(\frac{q_c}{P_{ssl}} + 1 \right)^{\frac{\gamma-1}{\gamma}} - 1 \right]} \quad (Eq \ 2.19) \quad 2.14$$

$$V_c = f(P_T - P_a) = f(q_c) \quad (Eq \ 2.20) \quad 2.14$$

$$V_{c_{Test}} = V_{c_{Std}} \quad (Eq \ 2.21) \quad 2.14$$

EQUATIONS

$$\frac{P'_T}{P_a} = \left[\frac{\gamma+1}{2} \left(\frac{V}{a} \right)^2 \right]^{\frac{\gamma}{\gamma-1}} \left[\frac{1}{\frac{2\gamma}{\gamma+1} \left(\frac{V}{a} \right)^2 - \frac{\gamma-1}{\gamma+1}} \right]^{\frac{1}{\gamma-1}}$$

(Eq 2.22) 2.15

$$\frac{q_c}{P_{ssl}} = \left[1 + 0.2 \left(\frac{V_c}{a_{ssl}} \right)^2 \right]^{3.5} - 1$$

(For $V_c \leq a_{ssl}$) (Eq 2.23) 2.15

$$\frac{q_c}{P_{ssl}} = \left[\frac{166.921 \left(\frac{V_c}{a_{ssl}} \right)^7}{\left[7 \left(\frac{V_c}{a_{ssl}} \right)^2 - 1 \right]^{2.5}} \right] - 1$$

(For $V_c \geq a_{ssl}$) (Eq 2.24) 2.15

$$V_e = \sqrt{\frac{2\gamma}{\gamma-1} \frac{P_a}{\rho_{ssl}} \left[\left(\frac{q_c}{P_a} + 1 \right)^{\frac{\gamma-1}{\gamma}} - 1 \right]}$$

(Eq 2.25) 2.17

$$V_e = V_T \sqrt{\sigma}$$

(Eq 2.26) 2.17

$$M = \frac{V_T}{a} = \frac{V_T}{\sqrt{\gamma g_c R T}} = \frac{V_T}{\sqrt{\gamma \frac{P}{\rho}}}$$

(Eq 2.27) 2.17

$$M = \sqrt{\frac{2}{\gamma-1} \left[\left(\frac{P_T - P_a}{P_a} + 1 \right)^{\frac{\gamma-1}{\gamma}} - 1 \right]}$$

(Eq 2.28) 2.17

$$\frac{P_T}{P_a} = \left(1 + \frac{\gamma-1}{2} M^2 \right)^{\frac{\gamma}{\gamma-1}}$$

(Eq 2.29) 2.18

FIXED WING PERFORMANCE

$$\frac{q_c}{P_a} = \left(1 + 0.2 M^2 \right)^{3.5} - 1 \quad \text{for } M < 1 \quad (\text{Eq 2.30}) \quad 2.18$$

$$\frac{q_c}{P_a} = \left[\frac{166.921 M^7}{\left(7M^2 - 1 \right)^{2.5}} \right] - 1 \quad \text{for } M > 1 \quad (\text{Eq 2.31}) \quad 2.18$$

$$M = f(P_T - P_a, P_a) = f(V_c, H_P) \quad (\text{Eq 2.32}) \quad 2.19$$

$$M_{\text{Test}} = M \quad (\text{Eq 2.33}) \quad 2.19$$

$$\Delta H_{P_{ic}} = H_{P_i} - H_{P_o} \quad (\text{Eq 2.34}) \quad 2.22$$

$$\Delta V_{ic} = V_i - V_o \quad (\text{Eq 2.35}) \quad 2.22$$

$$H_{P_i} = H_{P_o} + \Delta H_{P_{ic}} \quad (\text{Eq 2.36}) \quad 2.22$$

$$V_i = V_o + \Delta V_{ic} \quad (\text{Eq 2.37}) \quad 2.22$$

$$\Delta P = P_s - P_a \quad (\text{Eq 2.38}) \quad 2.27$$

$$\Delta V_{pos} = V_c - V_i \quad (\text{Eq 2.39}) \quad 2.27$$

$$\Delta H_{pos} = H_{P_c} - H_{P_i} \quad (\text{Eq 2.40}) \quad 2.27$$

$$\Delta M_{pos} = M - M_i \quad (\text{Eq 2.41}) \quad 2.27$$

$$\frac{P_s}{P_a} = f_1(M, \alpha, \beta, R_e) \quad (\text{Eq 2.42}) \quad 2.28$$

$$\frac{P_s}{P_a} = f_2(M, \alpha) \quad (\text{Eq 2.43}) \quad 2.28$$

$$\frac{\Delta P}{q_c} = f_3(M, \alpha) \quad (\text{Eq 2.44}) \quad 2.28$$

EQUATIONS

$$\frac{\Delta P}{q_c} = f_4(M) \text{ (High speed)} \quad (Eq\ 2.45) \quad 2.28$$

$$\frac{\Delta P}{q_c} = f_5(C_L) \text{ (Low speed)} \quad (Eq\ 2.46) \quad 2.28$$

$$\frac{\Delta P}{q_{c_i}} = f_6(M_i) \text{ (High speed)} \quad (Eq\ 2.47) \quad 2.29$$

$$\frac{\Delta P}{q_c} = f_7(W, V_c) \text{ (Low speed)} \quad (Eq\ 2.48) \quad 2.29$$

$$V_{c_W} = V_{c_{Test}} \sqrt{\frac{W_{Std}}{W_{Test}}} \quad (Eq\ 2.49) \quad 2.30$$

$$\frac{\Delta P}{q_c} = f_8(V_{c_W}) \text{ (Low speed)} \quad (Eq\ 2.50) \quad 2.30$$

$$V_{i_W} = V_{i_{Test}} \sqrt{\frac{W_{Std}}{W_{Test}}} \quad (Eq\ 2.51) \quad 2.30$$

$$\frac{\Delta P}{q_{c_i}} = f_9(V_{i_W}) \text{ (Low speed)} \quad (Eq\ 2.52) \quad 2.30$$

$$\frac{T_T}{T} = 1 + \frac{\gamma - 1}{2} M^2 \quad (Eq\ 2.53) \quad 2.32$$

$$\frac{T_T}{T} = 1 + \frac{\gamma - 1}{2} \frac{V_T^2}{\gamma g_c R T} \quad (Eq\ 2.54) \quad 2.32$$

$$\frac{T_T}{T} = 1 + \frac{K_T(\gamma - 1)}{2} M^2 \quad (Eq\ 2.55) \quad 2.33$$

$$\frac{T_T}{T} = 1 + \frac{K_T(\gamma - 1)}{2} \frac{V_T^2}{\gamma g_c R T} \quad (Eq\ 2.56) \quad 2.33$$

$$\frac{T_T}{T_a} = \frac{T_i}{T_a} = 1 + \frac{K_T M^2}{5} \quad (Eq\ 2.57) \quad 2.33$$

FIXED WING PERFORMANCE

$$T_T = T_i = T_a + \frac{K_T V_T^2}{7592} \quad (Eq\ 2.58) \quad 2.33$$

$$T_i = T_o + \Delta T_{ic} \quad (Eq\ 2.59) \quad 2.35$$

$$K_T = \left(\frac{T_i\ (^{\circ}K)}{T_a\ (^{\circ}K)} - 1 \right) \frac{5}{M^2} \quad (Eq\ 2.60) \quad 2.35$$

$$V_{G_1} = 3600 \left(\frac{D}{\Delta t_1} \right) \quad (Eq\ 2.61) \quad 2.50$$

$$V_{G_2} = 3600 \left(\frac{D}{\Delta t_2} \right) \quad (Eq\ 2.62) \quad 2.50$$

$$V_T = \frac{V_{G_1} + V_{G_2}}{2} \quad (Eq\ 2.63) \quad 2.50$$

$$\rho_a = \frac{P_a}{g_c R T_{a_ref}\ (^{\circ}K)} \quad (Eq\ 2.64) \quad 2.50$$

$$\sigma = \frac{\rho_a}{\rho_{ssl}} \quad (Eq\ 2.65) \quad 2.51$$

$$V_c = V_e - \Delta V_c \quad (Eq\ 2.66) \quad 2.51$$

$$M = \frac{V_T}{38.9678 \sqrt{T_{a_ref}\ (^{\circ}K)}} \quad (Eq\ 2.67) \quad 2.51$$

$$q_c = P_{ssl} \left\{ \left[1 + 0.2 \left(\frac{V_c}{a_{ssl}} \right)^2 \right]^{3.5} - 1 \right\} \quad (Eq\ 2.68) \quad 2.51$$

$$q_{c_i} = P_{ssl} \left\{ \left[1 + 0.2 \left(\frac{V_i}{a_{ssl}} \right)^2 \right]^{3.5} - 1 \right\} \quad (Eq\ 2.69) \quad 2.51$$

EQUATIONS

$$\Delta P = q_c - q_{c_i} \quad (Eq\ 2.70) \quad 2.51$$

$$V_{i_W} = V_i \sqrt{\frac{W_{Std}}{W_{Test}}} \quad (Eq\ 2.71) \quad 2.51$$

$$H_{P_{i_{ref}}} = H_{P_{o_{ref}}} + \Delta H_{P_{ic_{ref}}} \quad (Eq\ 2.72) \quad 2.54$$

$$H_{P_i} = \frac{T_{ssl}}{a_{ssl}} \left[1 - \left(\frac{P_s}{P_{ssl}} \right)^{\frac{1}{\left(\frac{g_{ssl}}{g_c a_{ssl} R} \right)}} \right] \quad (Eq\ 2.73) \quad 2.54$$

$$H_{P_{i_{ref}}} = \frac{T_{ssl}}{a_{ssl}} \left[1 - \left(\frac{P_a}{P_{ssl}} \right)^{\frac{1}{\left(\frac{g_{ssl}}{g_c a_{ssl} R} \right)}} \right] \quad (Eq\ 2.74) \quad 2.55$$

$$\Delta h = d \tan \theta \quad (Eq\ 2.75) \quad 2.57$$

$$\Delta h = L_{a/c} \frac{y}{x} \quad (Eq\ 2.76) \quad 2.57$$

$$H_{P_c} = H_{P_{c_{twr}}} + \Delta h \frac{T_{Std} (\text{°K})}{T_{Test} (\text{°K})} \quad (Eq\ 2.77) \quad 2.57$$

$$P_s = P_{ssl} \left(1 - 6.8755856 \times 10^{-6} H_{P_i} \right)^{5.255863} \quad (Eq\ 2.78) \quad 2.57$$

$$P_a = P_{ssl} \left(1 - 6.8755856 \times 10^{-6} H_{P_c} \right)^{5.255863} \quad (Eq\ 2.79) \quad 2.58$$

$$\text{Curve slope} = K_T \frac{\gamma - 1}{\gamma} T_a = 0.2 K_T T_a (\text{°K}) \text{ (High speed)} \quad (Eq\ 2.80) \quad 2.60$$

FIXED WING PERFORMANCE

$$\text{Curve slope} = K_T \frac{0.2 T_a (\text{°K})}{a_{ssl}^2} \text{ (Low speed)} \quad (Eq\ 2.81) \quad 2.60$$

$$K_T = \frac{\text{slope}}{0.2 T_a (\text{°K})} \text{ (High speed)} \quad (Eq\ 2.82) \quad 2.60$$

$$K_T = \frac{\text{slope } a_{ssl}^2}{0.2 T_a (\text{°K})} \text{ (Low speed)} \quad (Eq\ 2.83) \quad 2.61$$

$$M_i = \sqrt{\frac{2}{\gamma - 1} \left[\left(\frac{q_{c_i}}{P_s} + 1 \right)^{\frac{\gamma - 1}{\gamma}} - 1 \right]} \quad (Eq\ 2.84) \quad 2.62$$

$$\Delta P = \left(\frac{\Delta P}{q_{c_i}} \right) q_{c_i} \quad (Eq\ 2.85) \quad 2.62$$

$$q_c = q_{c_i} + \Delta P \quad (Eq\ 2.86) \quad 2.62$$

$$\Delta V_{pos} = V_c - V_{i_w} \quad (Eq\ 2.87) \quad 2.63$$

$$P_a = P_s - \Delta P \quad (Eq\ 2.88) \quad 2.63$$

$$H_{P_c} = \frac{T_{ssl}}{a_{ssl}} \left[1 - \left(\frac{P_a}{P_{ssl}} \right)^{\frac{1}{\left(\frac{g_{ssl}}{g_c a_{ssl} R} \right)}} \right] \quad (Eq\ 2.89) \quad 2.63$$

CHAPTER 3

$$C_{L_{max}(\Lambda)} = C_{L_{max}(\Lambda=0)} \cos(\Lambda) \quad (Eq\ 3.1) \quad 3.7$$

$$V_{e_s} = \sqrt{\frac{n_z W}{C_{L_s} q S}} \quad (Eq\ 3.2) \quad 3.10$$

EQUATIONS

$$\frac{V_{e_{s_1}}}{V_{e_{s_2}}} = \sqrt{\frac{C_{L_{s_2}}}{C_{L_{s_1}}}}$$
(Eq 3.3) 3.10

If $\frac{V_{e_{s_1}}}{V_{e_{s_2}}} = 0.8$, then $\frac{C_{L_{s_2}}}{C_{L_{s_1}}} = 0.64$

(Eq 3.4) 3.10

$$\frac{C_{L_{s_1}}}{C_{L_{s_2}}} = 1.56$$
(Eq 3.5) 3.10

$$\alpha_j = \alpha + \tau$$
(Eq 3.6) 3.17

$$L = L_{aero} + L_{Thrust}$$
(Eq 3.7) 3.17

$$L = n_z W$$
(Eq 3.8) 3.17

$$L_{Thrust} = T_G \sin \alpha_j$$
(Eq 3.9) 3.17

$$L = n_z W = L_{aero} + T_G \sin \alpha_j$$
(Eq 3.10) 3.17

$$C_L = \frac{L}{q S} = \frac{n_z W}{q S} = \frac{L_{aero}}{q S} + \frac{T_G \sin \alpha_j}{q S}$$
(Eq 3.11) 3.17

$$C_L = C_{L_{aero}} + \frac{T_G \sin \alpha_j}{q S}$$
(Eq 3.12) 3.17

$$C_L = C_{L_{aero}} + \frac{\frac{T_G \sin \alpha_j}{n_z W}}{C_L}$$
(Eq 3.13) 3.18

$$C_L = C_{L_{aero}} + C_L \left(\frac{T_G}{W} \frac{\sin \alpha_j}{n_z} \right)$$
(Eq 3.14) 3.18

$$C_L \left(1 - \frac{T_G}{W} \frac{\sin \alpha_j}{n_z} \right) = C_{L_{aero}} \quad (Eq\ 3.15) \quad 3.18$$

$$C_L = \frac{C_{L_{aero}}}{\left(1 - \frac{T_G}{W} \frac{\sin \alpha_j}{n_z} \right)} \quad (Eq\ 3.16) \quad 3.18$$

$$C_L = f \left(C_{L_{aero}}, \frac{T_G}{W}, \sin \alpha_j, n_z \right) \quad (Eq\ 3.17) \quad 3.18$$

$$C_{L_{aero}} = f(\alpha, M, R_e) \quad (Eq\ 3.18) \quad 3.19$$

$$C_{L_{aero}} = \frac{n_z W}{q S} = \frac{n_z W}{\frac{\gamma}{2} P_{ssl} S \delta M^2} \quad (Eq\ 3.19) \quad 3.19$$

$$C_{L_{aero}} = \frac{n_z \left(\frac{W}{\delta} \right)}{\left(\frac{\gamma}{2} P_{ssl} S \right) M^2} \quad (Eq\ 3.20) \quad 3.19$$

$$V_s = V_{s_{Test}} \sqrt{\frac{R+2}{R+1}} \text{ (For decelerations)} \quad (Eq\ 3.21) \quad 3.21$$

$$C_{L_s} = C_{L_{Test}} \left(\frac{R+1}{R+2} \right) \text{ (For decelerations)} \quad (Eq\ 3.22) \quad 3.21$$

$$V_s = V_{s_{Test}} \sqrt{\frac{R+1}{R+2}} \text{ (For accelerations)} \quad (Eq\ 3.23) \quad 3.21$$

$$R = \frac{V_{s_{Test}}}{\frac{c}{2} \dot{V}} \quad (Eq\ 3.24) \quad 3.21$$

$$C_{L_{max}} = C_{L_{max}} + K_d \left(\dot{V}_{Std} - \dot{V}_{Test} \right) \quad (Eq\ 3.25) \quad 3.22$$

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$$C_{L_{max}} = C_{L_{max}} + K_c (CG_{Std} - CG_{Test}) \quad (Eq\ 3.26) \quad 3.25$$

$$\Delta L_t (l_t) = T_G (Z) - D_R (Y) \quad (Eq\ 3.27) \quad 3.26$$

$$\Delta C_{L_t} = \frac{T_G}{q S} \left(\frac{Z}{l_t} \right) - \frac{D_R}{q S} \left(\frac{Y}{l_t} \right) \quad (Eq\ 3.28) \quad 3.26$$

$$R_e = \frac{\rho V c}{\mu} = V_e \sqrt{\rho} \left(\frac{\sqrt{\rho_{ssl}} c}{\mu} \right) \quad (Eq\ 3.29) \quad 3.27$$

$$C_{T_G} = \frac{T_G \sin \alpha_j}{q S} \quad (Eq\ 3.30) \quad 3.29$$

$$C_{L_{max}} = C_{L_{max}} + K_W (W_{Std} - W_{Test}) \quad (Eq\ 3.31) \quad 3.31$$

$$V_i = V_o + \Delta V_{ic} \quad (Eq\ 3.32) \quad 3.36$$

$$V_c = V_i + \Delta V_{pos} \quad (Eq\ 3.33) \quad 3.36$$

$$H_{P_i} = H_{P_o} + \Delta H_{P_{ic}} \quad (Eq\ 3.34) \quad 3.36$$

$$H_{P_c} = H_{P_i} + \Delta H_{pos} \quad (Eq\ 3.35) \quad 3.36$$

$$n_{z_i} = n_{z_o} + \Delta n_{z_{ic}} \quad (Eq\ 3.36) \quad 3.36$$

$$n_z = n_{z_i} + \Delta n_{z_{tare}} \quad (Eq\ 3.37) \quad 3.36$$

$$C_{L_{max}} = \frac{n_z W_{Test}}{0.7 P_{ssl} \delta_{Test} M^2 S} \quad (Eq\ 3.38) \quad 3.37$$

$$R = \frac{V_c}{\frac{c}{2} \dot{V}_{Test}} \quad (Eq\ 3.39) \quad 3.37$$

FIXED WING PERFORMANCE

$$\frac{C_{L_{max}}}{Std \dot{V}} = C_{L_{max}_{Test}} \left(\frac{R + 1}{R + 2} \right) \quad (Eq \ 3.40) \quad 3.37$$

$$\left(\frac{C_{L_{aero}}}{Std \dot{V}, CG, W} \right)_{Pwr \ ON} = \left(\frac{C_{L_{max}}}{Std \dot{V}, CG, W} \right)_{Pwr \ ON} - C_{T_G} \quad (Eq \ 3.41) \quad 3.42$$

$$\left(\frac{C_{L_{aero}}}{Std \dot{V}, CG, W} \right)_{Pwr \ OFF} = \left(\frac{C_{L_{aero}}}{Std \dot{V}, CG, W} \right)_{Pwr \ ON} - \Delta C_{L_t} - \Delta C_{L_E} \quad (Eq \ 3.42) \quad 3.42$$

$$V_{e_s} = \sqrt{\frac{841.5 n_z W}{C_{L_{max}} S}} \quad (Eq \ 3.43) \quad 3.44$$

CHAPTER 4

$$D_p = C_{D_p} q S \quad (Eq \ 4.1) \quad 4.4$$

$$D_i = L \alpha_i = C_L q S \alpha_i \quad (Eq \ 4.2) \quad 4.5$$

$$L = C_L q S \quad (Eq \ 4.3) \quad 4.5$$

$$D_i = C_{D_i} q S \quad (Eq \ 4.4) \quad 4.5$$

$$\alpha_i = \frac{C_L}{\pi A R} \quad (Eq \ 4.5) \quad 4.5$$

$$C_{D_i} = \frac{C_L^2}{\pi A R} \quad (Eq \ 4.6) \quad 4.6$$

$$C_{D_i} = \frac{C_L^2}{\pi e A R} \quad (Eq \ 4.7) \quad 4.6$$

$$D = D_p + D_i + D_M \quad (Eq \ 4.8) \quad 4.7$$

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$$C_D = C_{D_p} + C_{D_i} + C_{D_M} \quad (\text{Eq 4.9}) \quad 4.7$$

$$D = C_D qS \quad (\text{Eq 4.10}) \quad 4.8$$

$$C_D = C_{D_p} + \frac{C_L^2}{\pi e AR} \quad (\text{Eq 4.11}) \quad 4.8$$

$$D = C_{D_p} q S + \frac{C_L^2}{\pi e AR} q S \quad (\text{Eq 4.12}) \quad 4.8$$

$$L - W + T_G \sin \alpha_j = 0 \quad (\text{Eq 4.13}) \quad 4.9$$

$$L = W - T_G \sin \alpha_j \quad (\text{Eq 4.14}) \quad 4.9$$

$$C_L = \frac{L}{qS} \quad (\text{Eq 4.15}) \quad 4.9$$

$$C_L = \frac{W - T_G \sin \alpha_j}{qS} \quad (\text{Eq 4.16}) \quad 4.9$$

$$D = C_{D_p} qS + \frac{(W - T_G \sin \alpha_j)^2}{\pi e AR qS} \quad (\text{Eq 4.17}) \quad 4.9$$

$$D = C_{D_p} qS + \frac{W^2}{\pi e AR qS} \quad (\text{Eq 4.18}) \quad 4.9$$

$$q = \frac{1}{2} \rho_{ssl} V_e^2 \quad (\text{Eq 4.19}) \quad 4.9$$

$$q = \frac{1}{2} \rho_a V_T^2 \quad (\text{Eq 4.20}) \quad 4.9$$

$$q = \frac{1}{2} \gamma P_a M^2 \quad (\text{Eq 4.21}) \quad 4.10$$

FIXED WING PERFORMANCE

$$D = \frac{C_{D_p} \rho_{ssl} V_e^2 S}{2} + \frac{2 W^2}{\pi e AR S \rho_{ssl} V_e^2} \quad (Eq \ 4.22) \quad 4.10$$

$$D = \frac{C_{D_p} \rho_a V_T^2 S}{2} + \frac{2 W^2}{\pi e AR S \rho_a V_T^2} \quad (Eq \ 4.23) \quad 4.10$$

$$D = \frac{C_{D_p} \gamma P_a M^2 S}{2} + \frac{2 W^2}{\pi e AR S \gamma P_a M^2} \quad (Eq \ 4.24) \quad 4.10$$

$$D = \frac{C_{D_p(M)} \gamma P_a M^2 S}{2} + \frac{2 W^2}{\pi e_{(M)} AR S \gamma P_a M^2} \quad (Eq \ 4.25) \quad 4.12$$

$$\frac{D}{\delta} = \frac{C_{D_{P(M)}} \gamma P_{ssl} M^2 S}{2} + \frac{2 (W/\delta)^2}{\pi e_{(M)} AR S \gamma P_{ssl} M^2} \quad (Eq \ 4.26) \quad 4.12$$

$$\frac{D}{\delta} = f \left(M, \frac{W}{\delta} \right) \quad (Eq \ 4.27) \quad 4.13$$

$$\dot{W}_f = f \left(P, \rho, \mu, V, L, N \right) \quad (Eq \ 4.28) \quad 4.14$$

$$\frac{\dot{W}_f}{\delta \sqrt{\theta}} = f \left(M, \frac{N}{\sqrt{\theta}}, R_e \right) \quad (Eq \ 4.29) \quad 4.14$$

$$\frac{\dot{W}_f}{\delta \sqrt{\theta}} = f \left(M, \frac{N}{\sqrt{\theta}} \right) \quad (Eq \ 4.30) \quad 4.15$$

$$\frac{T_{N_x}}{\delta} = f \left(M, \frac{N}{\sqrt{\theta}} \right) \quad (Eq \ 4.31) \quad 4.15$$

EQUATIONS

$$\frac{\dot{W}_f}{\delta\sqrt{\theta}} = f \left(M, \frac{T_{N_x}}{\delta} \right) \quad (Eq \ 4.32) \quad 4.15$$

$$T_{N_x} = T_G \cos \alpha_j - T_R \quad (Eq \ 4.33) \quad 4.15$$

$$T_{N_x} = D \text{ (For small } \alpha_j, \text{ where } \cos \alpha_j \approx 1) \quad (Eq \ 4.34) \quad 4.15$$

$$\frac{T_{N_x}}{\delta} = \frac{D}{\delta} \quad (Eq \ 4.35) \quad 4.15$$

$$\frac{\dot{W}_f}{\delta\sqrt{\theta}} = f \left(M, \frac{D}{\delta} \right) \quad (Eq \ 4.36) \quad 4.15$$

$$\frac{\dot{W}_f}{\delta\sqrt{\theta}} = f \left(M, \frac{W}{\delta} \right) \quad (Eq \ 4.37) \quad 4.16$$

$$\theta = \frac{T_a}{T_{ssl}} \quad (Eq \ 4.38) \quad 4.21$$

$$\theta_T = \frac{T_T}{T_{ssl}} = \frac{OAT}{T_{ssl}} \quad (Eq \ 4.39) \quad 4.21$$

$$\frac{\dot{W}_f}{\delta\sqrt{\theta}} = f \left(M, OAT \right) \quad (Eq \ 4.40) \quad 4.22$$

$$T_T = T_a \left(1 + \frac{\gamma - 1}{2} M^2 \right) \quad (Eq \ 4.41) \quad 4.22$$

$$P_T = P_a \left(1 + \frac{\gamma - 1}{2} M^2 \right)^{\frac{\gamma}{\gamma - 1}} \quad (Eq \ 4.42) \quad 4.22$$

$$\delta = \frac{P_a}{P_{ssl}} \quad (Eq \ 4.43) \quad 4.23$$

FIXED WING PERFORMANCE

$$\delta_T = \frac{P_T}{P_{ssl}} \quad (Eq\ 4.44) \quad 4.23$$

$$\frac{\theta_T}{\theta} = (1 + 0.2 M^2) \quad (Eq\ 4.45) \quad 4.23$$

$$\frac{\delta_T}{\delta} = (1 + 0.2 M^2)^{3.5} \quad (Eq\ 4.46) \quad 4.23$$

$$\frac{\dot{W}_f}{\delta_T \sqrt{\theta_T}} = f(M, OAT) \quad (Eq\ 4.47) \quad 4.23$$

$$TSFC = \frac{\dot{W}_f}{T_{N_x}} \quad (Eq\ 4.48) \quad 4.25$$

$$T_{N_x} = D \quad (Eq\ 4.49) \quad 4.26$$

$$\dot{W}_f \approx D \quad (Eq\ 4.50) \quad 4.26$$

$$S.R. = \frac{nmi}{W_f} \quad (Eq\ 4.51) \quad 4.28$$

$$S.R. = \frac{V_T}{\dot{W}_f} \quad (Eq\ 4.52) \quad 4.28$$

$$S.E. = \frac{t}{W_{f_{Used}}} \quad (Eq\ 4.53) \quad 4.28$$

$$S.E. = \frac{1}{\dot{W}_f} \quad (Eq\ 4.54) \quad 4.29$$

$$Range = (S.R._{avg}) (Fuel\ Used) \quad (Eq\ 4.55) \quad 4.31$$

$$nmi = \frac{nmi}{lb} \times lb \quad (Eq\ 4.56) \quad 4.31$$

EQUATIONS

$$\text{THP} = \frac{T V_T}{550} = \frac{D V_T}{550} \quad (\text{Eq } 4.57) \quad 4.33$$

$$\text{THP} = \frac{C_D \rho_a V_T^3 S}{1100} + \frac{W^2}{275 \pi e A R S \rho_a V_T} \quad (\text{Eq } 4.58) \quad 4.33$$

$$\text{THP} = \text{THP}_p + \text{THP}_i \quad (\text{Eq } 4.59) \quad 4.34$$

$$D = \frac{C_D}{C_L} W \quad (\text{Eq } 4.60) \quad 4.35$$

$$\text{THP} = \frac{C_D W V_T}{C_L 550} \quad (\text{Eq } 4.61) \quad 4.35$$

$$L = W = C_L \frac{1}{2} \rho_a V_T^2 S \quad (\text{Eq } 4.62) \quad 4.35$$

$$\text{THP} = \frac{\left(\frac{2}{S}\right)^{\frac{1}{2}} W^{\frac{3}{2}} C_D}{\rho_a^{\frac{1}{2}} C_L^{\frac{3}{2}} 550} \quad (\text{Eq } 4.63) \quad 4.35$$

$$\rho_a = \rho_{ssl} \sigma \quad (\text{Eq } 4.64) \quad 4.35$$

$$\text{THP} = \frac{\sqrt{2} W^{\frac{3}{2}} C_D}{(S \rho_{ssl})^{\frac{1}{2}} \sqrt{\sigma} C_L^{\frac{3}{2}} 550} \quad (\text{Eq } 4.65) \quad 4.35$$

$$\text{THP}_e = \text{THP} \sqrt{\sigma} \quad (\text{Eq } 4.66) \quad 4.35$$

$$\text{THP}_e = \frac{\sqrt{2} W^{\frac{3}{2}} C_D}{(S \rho_{ssl})^{\frac{1}{2}} C_L^{\frac{3}{2}} 550} \quad (\text{Eq } 4.67) \quad 4.36$$

FIXED WING PERFORMANCE

$$THP = K_1 V_T^3 + K_2 V_T^{-1} \quad (Eq\ 4.68) \quad 4.37$$

$$3 K_1 V_T^2 - K_2 V_T^{-2} = 0 \quad (Eq\ 4.69) \quad 4.37$$

$$3 K_1 V_T^3 = K_2 V_T^{-1} \quad (Eq\ 4.70) \quad 4.37$$

$$3 THP_p = THP_i \quad (Eq\ 4.71) \quad 4.37$$

$$3 D_p = D_i \quad (Eq\ 4.72) \quad 4.37$$

$$3 C_{D_p} = C_{D_i} \quad (Eq\ 4.73) \quad 4.37$$

$$THP = K_o \frac{C_D}{\frac{3}{C_L^2}} \quad (Eq\ 4.74) \quad 4.38$$

$$V_T = \frac{V_e}{\sqrt{\sigma}} \quad (Eq\ 4.75) \quad 4.39$$

$$THP = \frac{THP_e}{\sqrt{\sigma}} \quad (Eq\ 4.76) \quad 4.39$$

$$\sigma = \frac{\rho_a}{\rho_{ssl}} \quad (Eq\ 4.77) \quad 4.40$$

$$THP_e V_e = \frac{C_{D_p} \rho_{ssl} V_e^4 S}{1100} + \frac{W^2}{275 \pi e AR S \rho_{ssl}} \quad (Eq\ 4.78) \quad 4.40$$

$$\eta_P = \frac{THP}{SHP} \quad (Eq\ 4.79) \quad 4.41$$

$$THP = \eta_P SHP \quad (Eq\ 4.80) \quad 4.41$$

$$THPSFC = \frac{\dot{W}_f}{THP} \quad (Eq\ 4.81) \quad 4.41$$

EQUATIONS

$$\text{SHPSFC} = \frac{\dot{W}_f}{\text{SHP}} \quad (Eq \ 4.82) \quad 4.42$$

$$\dot{W}_f = \frac{\text{THP}}{\eta_P} \text{ SHPSFC} \quad (Eq \ 4.83) \quad 4.42$$

$$\left(\frac{W}{\delta}\right)_{\text{Target}} = \frac{W + W_f}{\delta} \quad (Eq \ 4.84) \quad 4.47$$

$$\left(\frac{W}{\delta}\right)_{\text{Target}} = \frac{W_{\text{aircraft}} + W_f}{\delta} \quad (Eq \ 4.85) \quad 4.48$$

$$\delta = \frac{W_{\text{aircraft}}}{\left(\frac{W}{\delta}\right)_{\text{Target}}} + \frac{1}{\left(\frac{W}{\delta}\right)_{\text{Target}}} W_f \quad (Eq \ 4.86) \quad 4.48$$

$$OAT = T_a \left(1 + \frac{\gamma - 1}{2} K_T M^2 \right) \quad (Eq \ 4.87) \quad 4.48$$

$$H_{P_c} = H_{P_o} + \Delta H_{P_{ic}} + \Delta H_{\text{pos}} \quad (Eq \ 4.88) \quad 4.59$$

$$T_a = T_o + \Delta T_{ic} \quad (Eq \ 4.89) \quad 4.59$$

$$V_c = V_o + \Delta V_{ic} + \Delta V_{\text{pos}} \quad (Eq \ 4.90) \quad 4.59$$

$$V_T = \frac{V_c}{\sqrt{\sigma}} \quad (Eq \ 4.91) \quad 4.60$$

$$M = \frac{V_T}{a_{ssl} \sqrt{\theta}} \quad (Eq \ 4.92) \quad 4.60$$

$$\dot{W}_{f_{\text{ref}}} = \frac{\dot{W}_f}{\delta \sqrt{\theta}} \quad (Eq \ 4.93) \quad 4.60$$

$$W_{\text{ref}} = \frac{W}{\delta} \quad (Eq \ 4.94) \quad 4.60$$

FIXED WING PERFORMANCE

$$R_{Test} = \sum_{j=1}^n V_j \Delta t_j \quad (Eq\ 4.95) \quad 4.63$$

$$R.F._{Test} = \frac{t_T}{\ln \left(\frac{W_1}{W_2} \right)} \quad (Eq\ 4.96) \quad 4.63$$

$$R_{Std} = R.F._{Test} \ln \left(\frac{W_{Std_1}}{W_{Std_2}} \right) \quad (Eq\ 4.97) \quad 4.63$$

$$M = f(V_c, H_{P_c}) \quad (Eq\ 4.98) \quad 4.66$$

$$\delta = f(H_{P_c}) \quad (Eq\ 4.99) \quad 4.67$$

$$\frac{W + W_f}{\delta} = \frac{W}{\delta} \quad (Eq\ 4.100) \quad 4.67$$

$$\frac{W}{\delta} (\text{error}) = \frac{100 \left[\frac{W}{\delta} (\text{test}) - \frac{W}{\delta} (\text{target}) \right]}{\frac{W}{\delta} (\text{target})} \quad (Eq\ 4.101) \quad 4.67$$

$$^{\circ}\text{C} = ^{\circ}\text{K} - 273.15 \quad (Eq\ 4.102) \quad 4.67$$

$$OAT = f(T_a, M) \quad (Eq\ 4.103) \quad 4.67$$

$$T_a = f(OAT, M) \quad (Eq\ 4.104) \quad 4.67$$

$$S.R. \delta = \frac{661.483M}{\left(\frac{\dot{w}_f}{\delta \sqrt{\theta}} \right)} \quad (Eq\ 4.105) \quad 4.68$$

$$V_e = \sqrt{\frac{2q}{\rho_{ssl}}} = \sqrt{\frac{\sigma 2q}{\rho_a}} = \sqrt{\sigma} V_T \quad (Eq\ 4.106) \quad 4.73$$

$$SHP_e = SHP \sqrt{\sigma} \quad (Eq\ 4.107) \quad 4.73$$

EQUATIONS

$$T_a = T_{a_{\text{Std}}} + \Delta T_a \quad (\text{Eq 4.108}) \quad 4.78$$

$$\text{S.R.} = \frac{\frac{a_{\text{ssl}} M}{\dot{W}_f}}{\left(\frac{\dot{W}_f}{\delta \sqrt{\theta}}\right) \delta} \quad (\text{Eq 4.109}) \quad 4.78$$

$$V_{T_{\text{Hot day}}} = 661.483 M \sqrt{\theta_{\text{Std}}} \left(\frac{\sqrt{\theta_{\text{Hot day}}}}{\sqrt{\theta_{\text{Std}}}} \right) \quad (\text{Eq 4.110}) \quad 4.81$$

$$\dot{W}_{f_{\text{Hot day}}} = \dot{W}_{f_{\text{Std}}} \left(\frac{\sqrt{\theta_{\text{Hot day}}}}{\sqrt{\theta_{\text{Std}}}} \right) \quad (\text{Eq 4.111}) \quad 4.82$$

$$\frac{\dot{W}_f}{\sqrt{\theta}} = \varphi \quad (\text{Eq 4.112}) \quad 4.82$$

$$\text{Range} = \int_0^{W_{f_{\text{Used}}}} (\text{S.R.}) dW_f \quad (\text{Eq 4.113}) \quad 4.86$$

$$\text{Range} = \int_{W_1}^{W_2} (\text{S.R.}) dW \quad (\text{Eq 4.114}) \quad 4.87$$

$$\text{Range} = \int_{W_1}^{W_2} (\text{S.R.} \delta) \left(\frac{W}{\delta} \right) \frac{1}{W} dW \quad (\text{Eq 4.115}) \quad 4.87$$

$$\text{Range} = \int_{W_1}^{W_2} \left[\frac{661.483 M W}{\left(\frac{\dot{W}_f}{\delta \sqrt{\theta}} \right) \delta} \right] \frac{dW}{W} \quad (\text{Eq 4.116}) \quad 4.87$$

$$\frac{D}{\delta} = K_3 + K_4 \left(\frac{W}{\delta} \right)^2 \quad (\text{Eq 4.117}) \quad 4.88$$

FIXED WING PERFORMANCE

$$\text{R.F.} = \left[(\text{S.R. } \delta) \frac{W}{\delta} \right] \quad (\text{Eq 4.118}) \quad 4.90$$

$$\text{R.F.} = \left[\frac{661.483 M}{\left(\frac{\dot{W}_f}{\delta \sqrt{\theta}} \right)} \left(\frac{W}{\delta} \right) \right] \quad (\text{Eq 4.119}) \quad 4.90$$

$$\text{Range} = \int_{W_1}^{W_2} (\text{R.F.}) \frac{dW}{W} \quad (\text{Eq 4.120}) \quad 4.90$$

$$C_L = \frac{2 W}{\gamma P_a M^2 S} = \frac{2 \frac{W}{\delta}}{\gamma M^2 S P_{ssl}} \quad (\text{Eq 4.121}) \quad 4.101$$

$$C_L = f\left(\frac{W}{\delta}, M^2\right) \quad (\text{Eq 4.122}) \quad 4.101$$

$$\frac{\frac{W}{\delta}}{D} = \frac{W}{D} = \text{Constant} \quad (\text{Eq 4.123}) \quad 4.101$$

$$\frac{661.483 M}{\left(\frac{\dot{W}_f}{\delta \sqrt{\theta}} \right)} = \text{Constant} \quad (\text{Eq 4.124}) \quad 4.102$$

$$\frac{\text{TSFC}}{\sqrt{\theta}} = \text{Constant} \quad (\text{Eq 4.125}) \quad 4.102$$

$$\text{Range} = \text{R.F.} \ln \frac{W_1}{W_2} \quad (\text{Eq 4.126}) \quad 4.105$$

CHAPTER 5

$$TE = PE + KE \quad (Eq\ 5.1) \quad 5.2$$

$$PE = \int_0^h W dh \quad (Eq\ 5.2) \quad 5.2$$

$$PE = W \left(H_{P_c} + \Delta T \text{ correction} \right) \quad (Eq\ 5.3) \quad 5.2$$

$$KE = \frac{1}{2} \frac{W}{g} V_T^2 \quad (Eq\ 5.4) \quad 5.3$$

$$TE = W h + \frac{1}{2} \frac{W}{g} V_T^2 \quad (Eq\ 5.5) \quad 5.3$$

$$\frac{TE}{W} = h + \frac{V_T^2}{2g} \quad (Eq\ 5.6) \quad 5.3$$

$$E_h = h + \frac{V_T^2}{2g} \quad (Eq\ 5.7) \quad 5.3$$

$$\frac{d}{dt} E_h = \frac{d}{dt} \left(h + \frac{V_T^2}{2g} \right) \quad (Eq\ 5.8) \quad 5.7$$

$$\frac{d}{dt} E_h = \frac{dh}{dt} + \frac{V_T}{g} \frac{dV_T}{dt} \quad (Eq\ 5.9) \quad 5.7$$

$$\sum F_x = \frac{W}{g} \frac{dV_T}{dt} \quad (Eq\ 5.10) \quad 5.8$$

$$T_{N_x} = T_G \cos \alpha_j - T_R \quad (Eq\ 5.11) \quad 5.8$$

$$T_{N_x} - D - W \sin \gamma = \frac{W}{g} \frac{dV_T}{dt} \quad (Eq\ 5.12) \quad 5.8$$

$$\sin \gamma = \frac{V_T \text{ (vertical)}}{V_T \text{ (flight path)}} = \frac{\frac{dh}{dt}}{V_T} \quad (Eq\ 5.13) \quad 5.9$$

FIXED WING PERFORMANCE

$$T_{N_x} - D - W \frac{dh}{dt} \frac{1}{V_T} = \frac{W}{g} \frac{dV_T}{dt}$$

(Eq 5.14) 5.9

$$\frac{V_T (T_{N_x} - D)}{W} - \frac{dh}{dt} = \frac{V_T}{g} \frac{dV_T}{dt}$$

(Eq 5.15) 5.9

$$\frac{V_T (T_{N_x} - D)}{W} = \frac{dE_h}{dt}$$

(Eq 5.16) 5.9

$$P_s = \frac{V_T (T_{N_x} - D)}{W}$$

(Eq 5.17) 5.9

$$P_s = \frac{dE_h}{dt}$$

(Eq 5.18) 5.10

$$P_s = \frac{dh}{dt} + \frac{V_T}{g} \frac{dV_T}{dt}$$

(Eq 5.19) 5.10

$$P_s = \frac{V_T (T_{N_x} - D - \Delta D_i)}{W}$$

(Eq 5.20) 5.13

$$P_s = \frac{V_T (T_{N_x} - D - \Delta D_i)}{W + \Delta W}$$

(Eq 5.21) 5.15

$$P_s = \frac{V_T (T_{N_x} - D - \Delta D_p)}{W}$$

(Eq 5.22) 5.16

$$P_s = \frac{V_T (T_{N_x} + \Delta T_{N_x} - D)}{W}$$

(Eq 5.23) 5.17

$$P_s = \frac{(P_A - \Delta P_A) \cdot (P_{req} + \Delta P_{req})}{W}$$

(Eq 5.24) 5.19

EQUATIONS

$$q_c = P_{ssl} \left\{ \left[1 + 0.2 \left(\frac{V_c}{a_{ssl}} \right)^2 \right]^{3.5} - 1 \right\} \quad (Eq \ 5.25) \quad 5.28$$

$$P_a = P_{ssl} \left(1 - 6.8755856 \times 10^{-6} H_{P_c} \right)^{5.255863} \quad (Eq \ 5.26) \quad 5.28$$

$$M = \sqrt{\frac{2}{\gamma - 1} \left[\left(\frac{q_c}{P_a} + 1 \right)^{\frac{\gamma - 1}{\gamma}} - 1 \right]} \quad (Eq \ 5.27) \quad 5.28$$

$$T_a = \frac{OAT + 273.15}{1 + \frac{\gamma - 1}{2} K_T M^2} \quad (Eq \ 5.28) \quad 5.28$$

$$h = H_{P_c} \frac{T_{a_{Test}}}{T_{a_{Std}}} \quad (Eq \ 5.29) \quad 5.28$$

$$V_T = M \sqrt{\gamma g_c R T_a} \quad (Eq \ 5.30) \quad 5.28$$

$$\frac{W_{Test}}{W_{Std}} \quad (Eq \ 5.31) \quad 5.31$$

$$\frac{V_{T_{Std}}}{V_{T_{Test}}} = \frac{M_{Std} \sqrt{\theta_{Std}}}{M_{Test} \sqrt{\theta_{Test}}} \quad (Eq \ 5.32) \quad 5.32$$

$$\frac{V_{T_{Std}}}{V_{T_{Test}}} = \sqrt{\frac{T_{a_{Std}}}{T_{a_{Test}}}} \quad (Eq \ 5.33) \quad 5.32$$

$$\Delta T = f(T_a) \quad (Eq \ 5.34) \quad 5.32$$

$$\Delta D = D_{Std} - D_{Test} = \frac{2 \left(W_{Std}^2 - W_{Test}^2 \right)}{\pi e A R S \gamma P_a M^2} \quad (Eq \ 5.35) \quad 5.32$$

FIXED WING PERFORMANCE

$$P_{s_{Std}} = P_{s_{Test}} \frac{W_{Test}}{W_{Std}} \sqrt{\frac{T_{a_{Std}}}{T_{a_{Test}}}} + \frac{V_{T_{Std}}}{W_{Std}} (\Delta T_{N_x} - \Delta D) \quad (Eq\ 5.36) \quad 5.32$$

$$V_c = V_i + \Delta V_{pos} \quad (Eq\ 5.37) \quad 5.36$$

$$H_{P_c} = H_{P_i} + \Delta H_{pos} \quad (Eq\ 5.38) \quad 5.36$$

$$M = f(V_c, H_{P_c}) \quad (Eq\ 5.39) \quad 5.37$$

$$W_{Test} = \text{Initial } W - \int \dot{W}_f dt \quad (Eq\ 5.40) \quad 5.37$$

$$^{\circ}\text{C} = ^{\circ}\text{K} - 273.15 \quad (Eq\ 5.41) \quad 5.37$$

$$OAT = f(T_a, M_T) \quad (Eq\ 5.42) \quad 5.37$$

$$T_a = f(OAT, M) \quad (Eq\ 5.43) \quad 5.37$$

$$V_{T_{Test}} = f(V_c, H_{P_c}, T_a) \quad (Eq\ 5.44) \quad 5.37$$

$$V_{T_{Std}} = f(V_c, H_{P_c}, T_{Std}) \quad (Eq\ 5.45) \quad 5.37$$

$$h = H_{P_{c_{ref}}} + \Delta H_{P_c} \left(\frac{T_a}{T_{Std}} \right) \quad (Eq\ 5.46) \quad 5.37$$

$$E_h = h + \frac{V_{T_{Test}}^2}{2g} \quad (Eq\ 5.47) \quad 5.38$$

$$P_{s_{Test}} = \frac{dE_h}{dt} \quad (Eq\ 5.48) \quad 5.38$$

$$\gamma_{Test} = \sin^{-1} \left(\frac{dh/dt}{V_{T_{Test}}} \right) \quad (Eq\ 5.49) \quad 5.38$$

EQUATIONS

$$CCF = 1 + \left(\frac{V_{T_{Std}}}{g} \frac{dV}{dh} \right) \quad (Eq\ 5.50) \quad 5.38$$

$$P_{s_{Std}} = P_{s_{Test}} \left(\frac{W_{Test}}{W_{Std}} \right) \left(\frac{V_{T_{Std}}}{V_{T_{Test}}} \right) + \left(\frac{V_{T_{Std}}}{W_{Std}} \right) (\Delta T_{N_x} - \Delta D) \quad (Eq\ 5.51) \quad 5.38$$

$$\left(\frac{dh}{dt} \right)_{Std} = \frac{P_{s_{Std}}}{CCF} \quad (Eq\ 5.52) \quad 5.38$$

$$\gamma_{Std} = \sin^{-1} \left(\frac{\left(\frac{dh}{dt} \right)_{Std}}{V_{T_{Std}}} \right) \quad (Eq\ 5.53) \quad 5.39$$

$$|\gamma_{Test} - \gamma_{Std}| < 0.1 \quad (Eq\ 5.54) \quad 5.39$$

CHAPTER 6

$$n_z = \frac{L}{W} \quad (Eq\ 6.1) \quad 6.3$$

$$L^2 = W^2 + (W \tan \phi)^2 \quad (Eq\ 6.2) \quad 6.3$$

$$n_z^2 = 1 + \tan^2 \phi \quad (Eq\ 6.3) \quad 6.3$$

$$\tan \phi = \sqrt{(n_z^2 - 1)} \quad (Eq\ 6.4) \quad 6.4$$

$$L \cos \phi = W \quad (Eq\ 6.5) \quad 6.4$$

$$n_z = \frac{1}{\cos \phi} \quad (Eq\ 6.6) \quad 6.4$$

$$W \tan \phi = \frac{W}{g} a_R \quad (Eq\ 6.7) \quad 6.4$$

$$a_R = g \tan \phi \quad (Eq\ 6.8) \quad 6.4$$

FIXED WING PERFORMANCE

$$a_R = g \sqrt{\left(n_z^2 - 1 \right)}$$

(Eq 6.9) 6.4

$$R = \frac{V_T^2}{a_R}$$

(Eq 6.10) 6.5

$$R = \frac{V_T^2}{g \tan \phi}$$

(Eq 6.11) 6.5

$$R = \frac{V_T^2}{g \sqrt{\left(n_z^2 - 1 \right)}}$$

(Eq 6.12) 6.6

$$\omega = \frac{V_T}{R}$$

(Eq 6.13) 6.6

$$\omega = \frac{g}{V_T} \tan \phi$$

(Eq 6.14) 6.6

$$\omega = \frac{g}{V_T} \sqrt{\left(n_z^2 - 1 \right)}$$

(Eq 6.15) 6.6

$$\phi = \tan^{-1} \left(\frac{F_Y}{W} \right)$$

(Eq 6.16) 6.8

$$L = \frac{W}{\cos \phi}$$

(Eq 6.17) 6.8

$$\Delta L = F_Y \tan \phi$$

(Eq 6.18) 6.9

$$F_R = W \tan \phi + \frac{F_Y}{\cos \phi}$$

(Eq 6.19) 6.10

$$n_R = \tan \phi + \frac{n_Y}{\cos \phi}$$

(Eq 6.20) 6.10

$$\phi_E = \tan^{-1} \left(\tan \phi + \frac{n_Y}{\cos \phi} \right)$$

(Eq 6.21) 6.10

EQUATIONS

$$R = \frac{V_T^2}{g \left(\tan \phi + \frac{n_Y}{\cos \phi} \right)}$$
(Eq 6.22) 6.10

$$\omega = \frac{g \left(\tan \phi + \frac{n_Y}{\cos \phi} \right)}{V_T}$$
(Eq 6.23) 6.10

$$\Delta \omega = \frac{g n_Y}{V_T \cos \phi}$$
(Eq 6.24) 6.10

$$n_R = n_z - \cos \gamma$$
(Eq 6.25) 6.12

$$R_{(\text{wings level})} = \frac{V_T^2}{g (n_z - \cos \gamma)}$$
(Eq 6.26) 6.12

$$\omega_{(\text{wings level})} = \frac{g (n_z - \cos \gamma)}{V_T}$$
(Eq 6.27) 6.12

$$L = C_L q S + T_G \sin \alpha_j$$
(Eq 6.28) 6.16

$$n_z = \frac{C_L q}{W/S} + \frac{T_G}{W} \sin \alpha_j$$
(Eq 6.29) 6.16

$$n_{z_{\max}} = \frac{C_{L_{\max}} q}{(W/S)_{\min}}$$
(Eq 6.30) 6.17

$$n_{z_{\max}} = \frac{C_{L_{\max}}}{(W/S)_{\min}} 0.7 P_a M^2$$
(Eq 6.31) 6.17

$$n_{z_{\max}} = K M^2$$
(Eq 6.32) 6.17

$$K = \frac{0.7}{(W/S)} C_{L_{\max}} P_a$$
(Eq 6.33) 6.17

FIXED WING PERFORMANCE

$$\frac{1}{V_s^2 (1g)} = \frac{n_L}{V_A^2} \quad (Eq\ 6.34) \quad 6.18$$

$$V_A = V_s (1g) \sqrt{n_L} \quad (Eq\ 6.35) \quad 6.18$$

$$R = \frac{a^2 M^2}{g \sqrt{K^2 M^4 - 1}} \quad (Eq\ 6.36) \quad 6.20$$

$$\omega = \frac{g \sqrt{K^2 M^4 - 1}}{a M} \quad (Eq\ 6.37) \quad 6.21$$

$$K = \frac{0.7}{(W/S)} C_{L_{max}} P_a \quad (Eq\ 6.38) \quad 6.21$$

$$R_{min_{V>V_A}} = \left(\frac{a^2}{g \sqrt{n_L^2 - 1}} \right) M^2 \quad (Eq\ 6.39) \quad 6.23$$

$$\omega_{max_{V>V_A}} = \left(\frac{g \sqrt{n_L^2 - 1}}{a} \right) \frac{1}{M} \quad (Eq\ 6.40) \quad 6.23$$

$$D = \frac{C_D}{C_L} L \quad (Eq\ 6.41) \quad 6.29$$

$$T = \frac{C_D}{C_L} n_z W \quad (Eq\ 6.42) \quad 6.29$$

$$n_z = \frac{T}{W} \frac{C_L}{C_D} \quad (Eq\ 6.43) \quad 6.29$$

$$n_z_{sust_{max}} = \frac{T}{W} \left(\frac{C_L}{C_D} \right)_{max} \quad (Jet) \quad (Eq\ 6.44) \quad 6.29$$

EQUATIONS

$$n_z = \frac{T(V_T)}{W} \frac{L}{D(V_T)} \quad (Eq\ 6.45) \quad 6.29$$

$$n_{z_{sust_max}} = \frac{THP_{avail}}{W} \frac{L}{(THP_{req})_{min}} \quad (\text{Propeller}) \quad (Eq\ 6.46) \quad 6.30$$

$$\omega_{sust} = \frac{57.3 g}{V_T} \sqrt{n_{z_{sust}}^2 - 1} \quad (\text{deg/s}) \quad (Eq\ 6.47) \quad 6.30$$

$$R_{sust} = \frac{V_T^2}{g \sqrt{n_{z_{sust}}^2 - 1}} \quad (Eq\ 6.48) \quad 6.30$$

$$\frac{T}{\delta} = f \left(M, \frac{\dot{W}_f}{\delta_T \sqrt{\theta_T}} \right) \quad (Eq\ 6.49) \quad 6.32$$

$$\frac{\Delta D}{\delta} = \frac{\Delta T}{\delta} \quad (Eq\ 6.50) \quad 6.33$$

$$\Delta D_{Std-Test} = \frac{1}{\pi e AR S (0.7) P_{ssl} \delta_{Test} M^2} \left[(n_z W)_{Std}^2 - (n_z W)_{Test}^2 \right] \quad (Eq\ 6.51) \quad 6.34$$

$$n_{z_{Std}} = \sqrt{\frac{1}{W_{Std}^2} \left[(n_z W)_{Test}^2 + \Delta T \pi e AR (0.7) S P_{ssl} \delta_{Test} M^2 \right]} \quad (Eq\ 6.52) \quad 6.34$$

$$n_{z_{Std}} = n_{z_{Test}} \left(\frac{W_{Test}}{W_{Std}} \right) \quad (Eq\ 6.53) \quad 6.35$$

$$T_{ex} = T - D = \frac{W}{V_T} \frac{dh}{dt} + \frac{W}{g} \frac{dV_T}{dt} \quad (Eq\ 6.54) \quad 6.38$$

FIXED WING PERFORMANCE

$$\frac{dV_T}{dt} = \frac{11.3 P_s}{V_T} \quad (Eq\ 6.55) \quad 6.48$$

$$n_z = n_{z_0} + \Delta n_{ic} + \Delta n_{z_{tare}} \quad (Eq\ 6.56) \quad 6.57$$

$$V_i = V_o + \Delta V_{ic} \quad (Eq\ 6.57) \quad 6.63$$

$$V_c = V_i + \Delta V_{pos} \quad (Eq\ 6.58) \quad 6.63$$

$$H_{P_i} = H_{P_o} + \Delta H_{P_{ic}} \quad (Eq\ 6.59) \quad 6.63$$

$$H_{P_c} = H_{P_i} + \Delta H_{pos} \quad (Eq\ 6.60) \quad 6.63$$

$$n_{z_i} = n_{z_0} + \Delta n_{z_{ic}} \quad (Eq\ 6.61) \quad 6.63$$

$$n_{z_{Test}} = n_{z_i} + \Delta n_{z_{tare}} \quad (Eq\ 6.62) \quad 6.63$$

$$C_{L_{max_{Test}}} = \frac{n_{z_{Test}} W_{Test}}{0.7 P_{ssl} \delta_{Test} M^2 S} \quad (Eq\ 6.63) \quad 6.63$$

$$V_T = a M \quad (Eq\ 6.64) \quad 6.64$$

$$\Delta T = T_{Std} - T \quad (Eq\ 6.65) \quad 6.66$$

$$n_{z_{sust}} = \sqrt{\frac{P_{s_{1g}} \pi e AR S 0.7 P_{ssl} \delta M^2}{V_T W_{Std}}} + 1 \quad (Eq\ 6.66) \quad 6.68$$

$$T_{ex} = T - D = \frac{W_{Std}}{V_T} P_{s_{1g}} \quad (Eq\ 6.67) \quad 6.68$$

$$n_{z_{sust}} = \sqrt{\left(\frac{n_z W_{Std}}{\delta M}\right)^2 - \frac{\delta_h}{W_{Std}} M} \quad (Eq\ 6.68) \quad 6.68$$

EQUATIONS

$$n_z = \left(\frac{\delta}{W} \right) \left(0.7 P_{ssl} S \right) C_L M^2 \quad (Eq\ 6.69) \quad 6.71$$

$$\left(n_z \frac{W}{\delta} \right)_{Test} = \left(0.7 P_{ssl} S \right) C_L M^2 \quad (Eq\ 6.70) \quad 6.71$$

CHAPTER 7

$$E_h = h + \frac{V_T^2}{2g} \quad (Eq\ 7.1) \quad 7.2$$

$$P_s = \frac{dE}{dt} h = \frac{dh}{dt} + \frac{V_T}{g} \frac{dV_T}{dt} \quad (Eq\ 7.2) \quad 7.2$$

$$\frac{dV}{dt} = \frac{dV}{dh} \frac{dh}{dt} \quad (Eq\ 7.3) \quad 7.2$$

$$P_s = \frac{dh}{dt} + \frac{V}{g} \frac{dV}{dh} \frac{dh}{dt} \quad (Eq\ 7.4) \quad 7.3$$

$$P_s = \frac{dh}{dt} \left[1 + \frac{V}{g} \frac{dV}{dh} \right] \quad (Eq\ 7.5) \quad 7.3$$

$$\frac{dh}{dt} = P_s \left[\frac{1}{1 + \frac{V_T}{g} \frac{dV}{dh}} \right] \quad (Eq\ 7.6) \quad 7.3$$

$$CCF = \frac{1}{1 + \frac{V_T}{g} \frac{dV}{dh}} \quad (Eq\ 7.7) \quad 7.3$$

$$L - W \cos \gamma + T_G \sin \alpha_j = \frac{W}{g} a_z \quad (Eq\ 7.8) \quad 7.5$$

$$T_G \cos \alpha_j - T_R - D - W \sin \gamma = \frac{W}{g} a_x \quad (Eq\ 7.9) \quad 7.5$$

$$L = W \cos \gamma \quad (Eq\ 7.10) \quad 7.5$$

$$T_G \cos \alpha_j - T_R - D - W \sin \gamma = \frac{W}{g} \frac{dV_T}{dt} = 0 \quad (Eq\ 7.11) \quad 7.6$$

FIXED WING PERFORMANCE

$$T_{N_x} = T_G \cos \alpha_j - T_R \quad (Eq\ 7.12) \quad 7.6$$

$$T_{N_x} - D = W \sin \gamma \quad (Eq\ 7.13) \quad 7.6$$

$$\gamma = \sin^{-1} \left[\frac{T_{N_x} - D}{W} \right] \quad (Eq\ 7.14) \quad 7.6$$

$$V \sin \gamma = \frac{[T_{N_x} - D]}{W} V = \frac{dh}{dt} = V_v \quad (Eq\ 7.15) \quad 7.6$$

$$ROC = \frac{dh}{dt} = \frac{[T_{N_x} - D]}{W} V = \frac{T_{N_x} V - D V}{W} = \frac{P_A - P_{req}}{W} \quad (Eq\ 7.16) \quad 7.10$$

$$t = \int_0^h \frac{1}{dh} dh = \int_0^h \frac{1}{ROC} dh \quad (Eq\ 7.17) \quad 7.12$$

$$V_v = V_T \sin \gamma \quad (Eq\ 7.18) \quad 7.14$$

$$V_{hor} = V_T \cos \gamma \quad (Eq\ 7.19) \quad 7.14$$

$$\text{Rate of Climb} = P_s \left(\frac{1}{1 + \frac{V}{g} \frac{dV_T}{dh}} \right) \quad (Eq\ 7.20) \quad 7.20$$

$$\text{Time to Climb} = \int_{h_1}^{h_2} \frac{\left(1 + \frac{V}{g} \frac{dV_T}{dh} \right)}{P_s} dh \quad (Eq\ 7.21) \quad 7.20$$

$$\frac{dh}{dt} = \frac{dh + \frac{V}{g} dV_T}{P_s} \quad (Eq\ 7.22) \quad 7.21$$

$$dE_h = dh + \frac{V}{g} dV \quad (Eq\ 7.23) \quad 7.21$$

$$\frac{dE_h}{dt} = \frac{dE_h}{P_s} \quad (Eq\ 7.24) \quad 7.21$$

EQUATIONS

$$\text{Time to Climb} = \int_{E_{h_1}}^{E_{h_2}} \frac{1}{P_s} dE_h \quad (Eq\ 7.25) \quad 7.21$$

$$\frac{dE_h}{dW} = \frac{d}{dW} \left(h + \frac{V_T^2}{2g} \right) = \frac{dh}{dW} + \frac{V_T}{g} \frac{dV_T}{dW} \quad (Eq\ 7.26) \quad 7.25$$

$$\text{Fuel to Climb} = \int_{W_1}^{W_2} dW = - \int_{E_{h_1}}^{E_{h_2}} \frac{1}{\frac{dE_h}{dW}} dE_h \quad (Eq\ 7.27) \quad 7.25$$

$$-\frac{1}{\frac{dE_h}{dW}} = -\frac{dW}{dE_h} = -\frac{dW}{dt} \frac{dt}{dE_h} = \frac{\dot{W}_f}{P_s} \quad (Eq\ 7.28) \quad 7.25$$

$$\text{Fuel to Climb} = \int_{E_{h_1}}^{E_{h_2}} \frac{\dot{W}_f}{P_s} dE_h = \int_{E_{h_1}}^{E_{h_2}} \frac{1}{\frac{P_s}{\dot{W}_f}} dE_h \quad (Eq\ 7.29) \quad 7.26$$

$$E_{h_{Test}} = h_{Test} + \frac{V_{T_{Test}}^2}{2g} \quad (Eq\ 7.30) \quad 7.37$$

$$\left(\frac{dh}{dt} \right)_{Test} = P_{s_{Test}} \left(\frac{1}{1 + \frac{V_{T_{ref}}}{g} \frac{dV_T}{dh}} \right) \quad (Eq\ 7.31) \quad 7.38$$

$$P_{s_{Std}} = P_{s_{Test}} \frac{W_{Test}}{W_{Std}} \frac{V_{T_{Std}}}{V_{T_{Test}}} + \frac{V_{T_{Std}}}{W_{Std}} (\Delta T_{N_x} - \Delta D) \quad (Eq\ 7.32) \quad 7.38$$

$$\Delta D = D_{Std} - D_{Test} = \frac{2(W_{Std}^2 - W_{Test}^2)}{\pi e AR S \gamma P_a M^2} \quad (Eq\ 7.33) \quad 7.39$$

FIXED WING PERFORMANCE

$$L = n_z W \quad (Eq \ 7.34) \quad 7.40$$

$$n_z = \cos \gamma \quad (Eq \ 7.35) \quad 7.40$$

$$\Delta D = \frac{2 \left(W_{Std}^2 \cos^2 \gamma_{Std} - W_{Test}^2 \cos^2 \gamma_{Test} \right)}{\pi e AR \rho_{ssl} V_e^2 S} \quad (Eq \ 7.36) \quad 7.40$$

$$\Delta D = \frac{2 \left(W_{Std}^2 n_z^2 - W_{Test}^2 n_z^2 \right)}{\pi e AR \rho_{ssl} V_e^2 S} \quad (Eq \ 7.37) \quad 7.40$$

$$\left(\frac{dh}{dt} \right)_{Std} = P_{sStd} \left(\frac{1}{1 + \frac{V_T}{g} \frac{dV_T}{dh}} \right) \quad (Eq \ 7.38) \quad 7.41$$

$$\gamma_{Std} = \sin^{-1} \left(\frac{\frac{dh}{dt}}{V_T} \right) \quad (Eq \ 7.39) \quad 7.41$$

$$L = W \cos \gamma - T_G \sin \alpha_j \quad (Eq \ 7.40) \quad 7.42$$

$$n_z = \frac{L}{W} \quad (Eq \ 7.41) \quad 7.42$$

$$n_z = \cos \gamma - \frac{T_G}{W} \sin \alpha_j \quad (Eq \ 7.42) \quad 7.42$$

$$\Delta T_N = f(\Delta H_P, \Delta T_a) \quad (Eq \ 7.43) \quad 7.44$$

$$Distance = \int_{t_1}^{t_2} V_T \cos \gamma dt \quad (Eq \ 7.44) \quad 7.46$$

$$Fuel \ Used = \int_{t_1}^{t_2} \dot{W}_f dt \quad (Eq \ 7.45) \quad 7.47$$

$$^{\circ}C = ^{\circ}K - 273.15 \quad (Eq \ 7.46) \quad 7.52$$

EQUATIONS

$$OAT = f(T_a, M) \quad (Eq\ 7.47) \quad 7.52$$

$$T_a = f(OAT, M) \quad (Eq\ 7.48) \quad 7.52$$

$$V_T = f(OAT, M_T) \quad (Eq\ 7.49) \quad 7.53$$

$$h = H_{P_c\ ref} + \Delta H_{P_c} \left(\frac{T_a}{T_{std}} \right) \quad (Eq\ 7.50) \quad 7.53$$

CHAPTER 8

$$\sum F_z = L = W \cos \gamma \quad (Eq\ 8.1) \quad 8.3$$

$$\sum F_x = D = W \sin \gamma \quad (Eq\ 8.2) \quad 8.3$$

$$\frac{L}{D} = \frac{\cos \gamma}{\sin \gamma} = \cot \gamma \quad (Eq\ 8.3) \quad 8.3$$

$$V_{hor} = V_T \cos \gamma \quad (Eq\ 8.4) \quad 8.3$$

$$V_v = V_T \sin \gamma \quad (Eq\ 8.5) \quad 8.3$$

$$\frac{V_{hor}}{V_v} = \frac{V_T \cos \gamma}{V_T \sin \gamma} = \cot \gamma = \frac{L}{D} \quad (Eq\ 8.6) \quad 8.3$$

$$\sin \gamma = \frac{V_v}{V_T} \quad (Eq\ 8.7) \quad 8.3$$

$$\gamma = \sin^{-1} \left(\frac{V_v}{V_T} \right) \quad (Eq\ 8.8) \quad 8.4$$

$$\gamma = \sin^{-1} \left(\frac{dh/dt}{V_T} \right) \quad (Eq\ 8.9) \quad 8.4$$

$$\frac{L}{D} = \cot \left[\sin^{-1} \left(\frac{dh/dt}{V_T} \right) \right] \quad (Eq\ 8.10) \quad 8.4$$

FIXED WING PERFORMANCE

$$\text{Glide Ratio} = \frac{L}{D} = \frac{V_T \cos \gamma}{V_v}$$

(Eq 8.11) 8.4

$$GR = \frac{L}{D} \approx \frac{V_T}{V_v}$$

(Eq 8.12) 8.4

$$V_T = \sqrt{V_T^2 \sin^2 \gamma + V_T^2 \cos^2 \gamma} = \sqrt{V_T^2 (\sin^2 \gamma + \cos^2 \gamma)}$$

(Eq 8.13) 8.5

$$\sum F_x = W \sin \gamma - D = \frac{W}{g} \frac{dV_T}{dt}$$

(Eq 8.14) 8.10

$$\frac{D}{W} = \sin \gamma - \frac{1}{g} \frac{dV_T}{dt}$$

(Eq 8.15) 8.10

$$\frac{dV_T}{dt} = \frac{dV_T}{dh} \frac{dh}{dt}$$

(Eq 8.16) 8.10

$$\frac{D}{W} = \sin \gamma - \frac{1}{g} \frac{dV_T}{dh} \frac{dh}{dt}$$

(Eq 8.17) 8.10

$$\frac{dh}{dt} = V_T \sin \gamma$$

(Eq 8.18) 8.11

$$\frac{D}{W} = \sin \gamma - \frac{1}{g} \frac{dV_T}{dh} V_T \sin \gamma$$

(Eq 8.19) 8.11

$$\frac{L}{W} = \cos \gamma$$

(Eq 8.20) 8.11

$$\frac{L}{D} = \cot \gamma \left[\frac{1}{1 - \frac{V_T}{g} \frac{dV_T}{dh}} \right]$$

(Eq 8.21) 8.11

$$\frac{L}{D} = \cot \left[\sin^{-1} \left(\frac{(dh/dt)}{V_T} \right) \right] \left[\frac{1}{1 - \frac{V_T}{g} \frac{dV_T}{dh}} \right]$$

(Eq 8.22) 8.11

$$\frac{L}{D} = \left[\frac{GR}{1 - \frac{V_T}{g} \frac{dV_T}{dh}} \right]$$

(Eq 8.23) 8.12

EQUATIONS

$$GR = \frac{L}{D} \left[1 - \frac{V_T}{g} \frac{dV_T}{dh} \right] \quad (Eq\ 8.24) \quad 8.12$$

$$V_c = V_i + \Delta V_{pos} \quad (Eq\ 8.25) \quad 8.29$$

$$H_{P_c} = H_{P_i} + \Delta H_{pos} \quad (Eq\ 8.26) \quad 8.29$$

$$T_a (\text{ }^{\circ}\text{C}) = T_a (\text{ }^{\circ}\text{K}) - 273.15 \quad (Eq\ 8.27) \quad 8.29$$

$$OAT = f(T_a, M_T) \quad (Eq\ 8.28) \quad 8.29$$

$$T_a = f(OAT, M_T) \quad (Eq\ 8.29) \quad 8.29$$

$$V_{T_{Test}} = f(V_c, H_{P_c}, T_a) \quad (Eq\ 8.30) \quad 8.29$$

$$V_{T_{Std}} = f(V_c, H_{P_c}, T_{Std}) \quad (Eq\ 8.31) \quad 8.29$$

$$h = H_{P_c \text{ ref}} + \Delta H_{P_c} \left(\frac{T_a}{T_{Std}} \right) \quad (Eq\ 8.32) \quad 8.29$$

$$E_h = h + \frac{V_{T_{Test}}^2}{2g} \quad (Eq\ 8.33) \quad 8.30$$

$$P_{s_{Test}} = \frac{dE_h}{dt} \quad (Eq\ 8.34) \quad 8.30$$

$$P_{s_{Std}} = P_{s_{Test}} \left(\frac{W_{Test}}{W_{Std}} \right) \left(\frac{V_{T_{Std}}}{V_{T_{Test}}} \right) + \left(\frac{V_{T_{Std}}}{W_{Std}} \right) (\Delta T_{N_x} - \Delta D) \quad (Eq\ 8.35) \quad 8.30$$

$$\gamma_{Test} = \sin^{-1} \left[\frac{dh/dt}{V_{T_{Test}}} \right] \quad (Eq\ 8.36) \quad 8.30$$

FIXED WING PERFORMANCE

$$DCF = 1 + \left(\frac{V_{T_{Std}}}{g} \frac{dV_T}{dh} \right) \quad (Eq\ 8.37) \quad 8.30$$

$$\left(\frac{dh}{dt} \right)_{Std} = \frac{P_{s_{Std}}}{DCF} \quad (Eq\ 8.38) \quad 8.30$$

$$\gamma_{Std} = \sin^{-1} \left[\frac{(dh/dt)_{Std}}{V_{T_{Std}}} \right] \quad (Eq\ 8.39) \quad 8.30$$

$$V_i = V_o + \Delta V_{ic} \quad (Eq\ 8.40) \quad 8.32$$

$$H_{P_i} = H_{P_o} + \Delta H_{P_{ic}} \quad (Eq\ 8.41) \quad 8.32$$

$$T_i = T_o + \Delta T_{ic} \quad (Eq\ 8.42) \quad 8.32$$

$$V_T = 39.0 M \sqrt{T_a (\text{°K})} \quad (Eq\ 8.43) \quad 8.32$$

$$W_{f_{Used}} = W_{f_{Start}} - W_{f_{End}} \quad (Eq\ 8.44) \quad 8.32$$

$$\Delta d = V_{T_{avg}} \frac{\Delta t}{60} \quad (Eq\ 8.45) \quad 8.32$$

$$Range = \sum_{Sea\ Level}^{H_p} \Delta d \quad (Eq\ 8.46) \quad 8.32$$

EQUATIONS

CHAPTER 9

$$R = \mu (W - L) \quad (\text{Eq 9.1}) \quad 9.5$$

$$\int_0^1 [T - D - \mu(W - L)] dS = \frac{1}{2} \frac{W}{g} \left(V_{TO}^2 \right) \quad (\text{Eq 9.2}) \quad 9.6$$

$$[T - D - \mu(W - L)]_{\text{Avg}} S_1 = \frac{1}{2} \frac{W}{g} \left(V_{TO}^2 \right) \quad (\text{Eq 9.3}) \quad 9.6$$

$$S_1 = \frac{W V_{TO}^2}{2g [T - D - \mu(W - L)]_{\text{Avg}}} \quad (\text{Eq 9.4}) \quad 9.7$$

$$\text{Work} = \Delta T V \Delta t \quad (\text{Eq 9.5}) \quad 9.9$$

$$T_{ex} = T - D - \mu(W - L) \quad (\text{Eq 9.6}) \quad 9.9$$

$$q = \frac{1}{2} \rho V^2 \quad (\text{Eq 9.7}) \quad 9.9$$

$$D = C_D q S \quad (\text{Eq 9.8}) \quad 9.9$$

$$L = C_L q S \quad (\text{Eq 9.9}) \quad 9.9$$

$$C_D = C_{D_p} + C_{D_i} \quad (\text{Eq 9.10}) \quad 9.9$$

$$C_{D_i} = \frac{C_L^2}{\pi e A R} \quad (\text{Eq 9.11}) \quad 9.10$$

$$C_D = C_{D_p} + \frac{C_L^2}{\pi e A R} \quad (\text{Eq 9.12}) \quad 9.10$$

$$D = \left(C_{D_p} + \frac{C_L^2}{\pi e A R} \right) q S \quad (\text{Eq 9.13}) \quad 9.10$$

FIXED WING PERFORMANCE

$$T_{ex} = T - \left(C_{D_p} + \frac{C_L^2}{\pi e AR} \right) q S - \mu (W - C_L q S) \quad (Eq \ 9.14) \quad 9.10$$

$$\frac{dT_{ex}}{dC_L} = \left(\frac{2 C_L}{\pi e AR} \right) q S + \mu (q S) \quad (Eq \ 9.15) \quad 9.10$$

$$C_{L_{Opt}} = \frac{\mu \pi e AR}{2} \quad (Eq \ 9.16) \quad 9.10$$

$$S_2 = \int_{\text{Lift off}}^{50 \text{ ft}} (T - D) dS = \frac{W}{2g} \left(V_{50}^2 - V_{TO}^2 \right) + 50 W \quad (Eq \ 9.17) \quad 9.12$$

$$S_2 = \frac{W \left(\frac{V_{50}^2 - V_{TO}^2}{2g} + 50 \right)}{(T - D)_{Avg}} \quad (Eq \ 9.18) \quad 9.12$$

$$V_{TO_w} = V_{TO} - V_w \quad (Eq \ 9.19) \quad 9.13$$

$$S_{1w} = \frac{W V_{TO_w}^2}{2 g T_{ex,Avg_w}} \quad (Eq \ 9.20) \quad 9.13$$

$$S_{1std} = \frac{W \left(V_{TO_w} + V_w \right)^2}{2 g T_{ex,Avg}} \quad (Eq \ 9.21) \quad 9.13$$

$$S_{1std} = S_{1w} \frac{T_{ex,Avg_w}}{T_{ex,Avg}} \left(1 + \frac{V_w}{V_{TO_w}} \right)^2 \quad (Eq \ 9.22) \quad 9.13$$

$$S_{1std} = S_{1w} \left(1 + \frac{V_w}{V_{TO_w}} \right)^{1.85} \quad (Eq \ 9.23) \quad 9.14$$

EQUATIONS

$$S_{2_{\text{Std}}} = S_{2_w} + \Delta S_2 \quad (\text{Eq 9.24}) \quad 9.14$$

$$T_{\text{ex}_{\text{Avg}}} S_{1_{\text{SL}}} = \frac{1}{2} \frac{W}{g} V_{\text{TO}}^2 - W S_{1_{\text{SL}}} \sin \theta \quad (\text{Eq 9.25}) \quad 9.15$$

$$S_{1_{\text{SL}}} = \frac{W V_{\text{TO}}^2}{2 g \left(T_{\text{ex}_{\text{Avg}}} + W \sin \theta \right)} \quad (\text{Eq 9.26}) \quad 9.15$$

$$S_{1_{\text{Std}}} = \frac{S_{1_{\text{SL}}}}{\left(1 - \frac{2g S_{1_{\text{SL}}}}{V_{\text{TO}}^2} \sin \theta \right)} \quad (\text{Eq 9.27}) \quad 9.15$$

$$S_{1_{\text{Std}}} = S_{1_{\text{Test}}} \left(\frac{W_{\text{Std}}}{W_{\text{Test}}} \right)^{2.3} \left(\frac{\sigma_{\text{Test}}}{\sigma_{\text{Std}}} \right) \left(\frac{T_{N_{\text{Test}}}}{T_{N_{\text{Std}}}} \right)^{1.3} \quad (\text{Eq 9.28}) \quad 9.16$$

$$S_{2_{\text{Std}}} = S_{2_{\text{Test}}} \left(\frac{W_{\text{Std}}}{W_{\text{Test}}} \right)^{2.3} \left(\frac{\sigma_{\text{Test}}}{\sigma_{\text{Std}}} \right)^{0.7} \left(\frac{T_{N_{\text{Test}}}}{T_{N_{\text{Std}}}} \right)^{1.6} \quad (\text{Eq 9.29}) \quad 9.16$$

$$S_{1_{\text{Std}}} = S_{1_{\text{Test}}} \left(\frac{W_{\text{Std}}}{W_{\text{Test}}} \right)^{2.6} \left(\frac{\sigma_{\text{Test}}}{\sigma_{\text{Std}}} \right)^{1.9} \left(\frac{N_{\text{Test}}}{N_{\text{Std}}} \right)^{0.7} \left(\frac{P_{a_{\text{Test}}}}{P_{a_{\text{Std}}}} \right)^{0.5} \quad (\text{Eq 9.30}) \quad 9.17$$

$$S_{2_{\text{Std}}} = S_{2_{\text{Test}}} \left(\frac{W_{\text{Std}}}{W_{\text{Test}}} \right)^{2.6} \left(\frac{\sigma_{\text{Test}}}{\sigma_{\text{Std}}} \right)^{1.9} \left(\frac{N_{\text{Test}}}{N_{\text{Std}}} \right)^{0.8} \left(\frac{P_{a_{\text{Test}}}}{P_{a_{\text{Std}}}} \right)^{0.6} \quad (\text{Eq 9.31}) \quad 9.17$$

$$S_3 = \frac{W \left(\frac{V_{\text{TD}}^2 - V_{50}^2}{2g} - 50 \right)}{(T - D)_{\text{Avg}}} \quad (\text{Eq 9.32}) \quad 9.19$$

FIXED WING PERFORMANCE

$$S_4 = \int_{\text{Touchdown}}^{\text{Stop}} [T - D - \mu(W - L)] dS = \frac{1}{2} \frac{W}{g} \left(0 - V_{TD}^2 \right) \quad (\text{Eq 9.33}) \quad 9.19$$

$$S_4 = \frac{-W V_{TD}^2}{2g [T - D - \mu(W - L)]_{\text{Avg}}} \quad (\text{Eq 9.34}) \quad 9.20$$

$$S_{3_{\text{Std}}} = S_{3_{\text{Test}}} \left(\frac{W_{\text{Std}}}{W_{\text{Test}}} \right)^{\left(2 + \frac{E_h}{E_h + 50} \right)} \left(\frac{\sigma_{\text{Test}}}{\sigma_{\text{Std}}} \right)^{\left(\frac{E_h}{E_h + 50} \right)} \quad (\text{Eq 9.35}) \quad 9.23$$

$$E_h = \frac{V_{50}^2 - V_{TD}^2}{2g} \quad (\text{Eq 9.36}) \quad 9.23$$

$$S_{4_{\text{Std}}} = S_{4_{\text{Test}}} \left(\frac{W_{\text{Std}}}{W_{\text{Test}}} \right)^2 \left(\frac{\sigma_{\text{Test}}}{\sigma_{\text{Std}}} \right) \quad (\text{Eq 9.37}) \quad 9.23$$

$$V_w = \text{Wind Velocity} \cos(\text{Wind Direction Relative To Runway}) \quad (\text{Eq 9.38}) \quad 9.28$$

$$\sigma = 9.625 \frac{P_a}{T_a} \quad (\text{Eq 9.39}) \quad 9.29$$

$$V_{TD_w} = V_{TD} - V_w \quad (\text{Eq 9.40}) \quad 9.30$$

$$S_{4_{\text{Std}}} = S_{4_w} \left(1 + \frac{V_w}{V_{TD}} \right)^{1.85} \quad (\text{Eq 9.41}) \quad 9.30$$

$$S_{4_{\text{Std}}} = \frac{S_{4_{\text{SL}}}}{\left(1 - \frac{2g S_{4_{\text{SL}}}}{V_{TD}^2} \sin \theta \right)} \quad (\text{Eq 9.42}) \quad 9.31$$